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NACA airfoil - Wikipedia

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NACA 4 digit Airfoil: Nomenclature and Equations | The ...

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The NACA five-digit series describes more complex airfoil shapes: [6] The first digit, when multiplied by 0.15, gives the designed coefficient of lift (C L) . Second and third digits, when divided by 2, give p , the distance of maximum camber from the leading edge (as per cent of chord).

NACA airfoil

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The NACA airfoil series The early NACA airfoil series, the 4-digit, 5-digit, and modified 4-/5-digit, were generated using analytical equations that describe the camber (curvature) of the mean-line (geometric centerline) of the airfoil section as well as the section's thickness distribution along the length of the airfoil. Later

The NACA airfoil series - Stanford University

The NACA airfoil section is created from a camber line and a thickness distribution plotted perpendicular to the camber line. The equation for the camber line is split into sections either side of the point of maximum camber position (P). In order to calculate the position of the final airfoil envelope later the gradient of the camber line is also required. The equations are:

NACA 4 digit airfoil generator (NACA 2412 AIRFOIL)

Airfoil database search (NACA 6 series) Search the 1638 airfoils available in the databases filtering by name, thickness and camber. Click on an airfoil image to display a larger preview picture. There are links to the original airfoil source and dat file and the details page with polar diagrams for a range of Reynolds numbers.

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National Advisory Committee for Aeronautics airfoils. During the late 1920s and into the 1930s, the NACA developed a series of thoroughly tested airfoils and devised a numerical designation for each airfoil — a four digit number that represented the airfoil section's critical geometric properties.

NACA Airfoils | NASA

NACA 0012 airfoil Max thickness 12% at 30% chord Max camber 0% at 0% chord Source UIUC Airfoil Coordinates Database (n2414-il) NACA 2414: Airfoil details Send to airfoil plotter Add to comparison Lednicer format dat file Selig format dat file Source dat file: NACA 2414 airfoil Max thickness 14% at 29.5% chord Max camber 2% at 39.6% chord

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Naca Airfoil Data - Target Telecoms

The NACA 6a-series airfoil sections were designed to eliminate the trailing-edge cusp which is characteristic of the NACA 6a-series sections. Theoretical data are presented for NACA 6a-series basic thickness forms having the position of minimum pressure of 30, 40, and 50 percent chord and with thickness ratios varying from 6 percent to 15 percent.